NASA SBIR 2005 Phase I Solicitation

O2 Space Transportation

Achieving space flight can be astonishing. It is an undertaking of great complexity, requiring numerous technological and engineering disciplines and a high level of organizational skill. Overcoming Earth's gravity to achieve orbit demands collections of quality data to maintain the security required of the range. The harsh environment of space puts tight constraints on the equipment needed to perform the necessary functions. Not only is there a concern for safety but the 2004 Space Transportation Policy directive states that the US maintain a robust transportation capabilities to assure access to space. Given this backdrop, this topic is designed to address technologies to enable a safer and more reliable space transportation capability. Paramount to obtaining these capabilities is the development of the technologies identified in the following subtopics.

Subtopics

O2.01 Automated Optical Tracking and Identification of 3D Tumbling Objects

Lead Center: KSC
Participating Center(s): GSFC

Automated Optical Tracking

Develop a fully automated optical tracking system using data from multiple tracking stations located in and around the spaceport to provide accurate real-time trajectory and range data on space-lift vehicles for as long as possible following launch. The necessary optical tracking algorithms will be developed and modeled, with an emphasis on a robust automated tracking capability in the presence of smoke, clouds, or haze. The camera locations may be either land or sea based, or mounted on aerial vehicles, or some combination of all three. The initial investigation will determine the maximum downrange tracking distance, the tracking errors as a function of downrange distance, the processing speed, and the means for transmitting analyzed data to the command center.

Tracking and Identification of 3D Tumbling Objects

Develop techniques to track and construct 3-dimensional views of tumbling objects in the atmosphere or space using digital optical tracking images for a variety of missions. These views will be used to determine the objects' approximate geometric sizes and shapes. The potential application is to help track and identify debris quickly after an accident or flight anomaly. The data will be provided by sequential digital images from one or more tracking cameras, ideally operating autonomously. The goal is to track and identify between 50 to 100 objects with typical cross-sections varying from tens of square meters down to one square meter or less within several minutes after an
accident. The initial investigation will determine the minimum size that can be imaged using current technology, the probability of correctly estimating an object's size and shape, the processing speed, and the means for transmitting analyzed data to the command center.

O2.02 Space Transportation Test Requirements and Instrumentation

Lead Center: SSC
Participating Center(s): GRC, MSFC

Ground testing of propulsion systems is a critical requirement to enable NASA's New Vision for Exploration announced by President Bush. Relevant ground testing technologies and capabilities are crucial to the Development, Qualification, and Acceptance process of validating cargo launch vehicles and Human Rated Vehicles including Crew Exploration Vehicles (CEV), CEV Launch Systems, Cargo Launch Systems, and Lunar Surface Access Modules propulsion systems. The ability to quickly and efficiently perform system certification greatly impacts all space programs. While hydrocarbon engines have not been officially selected as the next generation of launch vehicle propulsion systems, it is widely agreed that a return to hydrocarbon propellants is necessary to achieve the heavy lift capability required by these programs.

Engine Plume Sensors and Modeling

Engine plume in situ remote measurements during ground testing can provide information on the engine health and the engine performance as well as concentrations of environmentally sensitive species without disturbing the engine testing. In particular, since hydrocarbon fueled rocket engine plumes contain carbon soot because of fuel-rich combustion, soot measurements and modeling are very important. Rocket plume sensors must be able to detect gas species, temperature, and velocity for hydrocarbons (kerosene), and hybrid fuels.

Innovative sensors are required for measuring flow rate, temperature, pressure, rocket engine plume constituents, and effluent gas detection. Sensors must not physically intrude into the measurement space. High accuracy (0.2%), low-millisecond to sub-millisecond response time is required. Temperature sensors must be able to measure both cryogenic and high temperature fluids under high pressure (up to 15,000 psi) and high flow rate conditions (82 ft/sec). Response time must be on the order of a few milliseconds to sub-milliseconds.

Innovative methods in phenomenology, modeling, sensors, and instrumentation for the prediction, characterization, and measurement of rocket engine combustion instability are of interest. Sensor systems should have bandwidth capabilities in excess of 100 kHz. Emphasis is on development of optical-based sensor systems that will be non-intrusive in the test article hardware or exhaust plume.

Sensors must support integration with Integrated Systems Health Management (ISHM) technologies.

Engine Acoustic Energy Prediction and Sensing

The high levels of acoustic energy generated by rocket engines can be destructive to both launch vehicles and
ground test/launch facilities. Current acoustic energy prediction methods can only provide a rough order of
magnitude estimate of the amount of acoustic energy that a rocket engine will generate. Consequently, facilities
and vehicles may be unnecessarily over designed to withstand the higher predictions, adding unnecessary weight
and complexity.

New and innovative acoustic measurement techniques and sensors are necessary to accurately measure and
predict the rocket plume acoustic environment. Current methods of predicting far-field and near-field acoustic levels
produced by rocket engines rely on empirical models and require numerous physical measurements. New and
innovative acoustic prediction methods are required that can accurately predict the acoustic levels a priori or using
fewer measurements. New, innovative techniques based on energy density measurements rather than pressure
measurements show promise as replacements for the older models.

Computational and Modeling Tools and Methods

The wide range of pressures, flow rates, and temperatures associated with rocket propulsion systems result in
complex dynamics. It is not realistic to physically test each component and the component-to-component
interaction in all states before designing a system. Consequently, systems must be tuned after fabrication, requiring
extensive testing and often component redesign. Tools and methods are necessary that will allow the use of
computational methods to accurately model and predict system performance.

Development of tools that integrate simple operator interfaces with detailed design and/or analysis software for
modeling and enhancing the flow performance of flow system components such as valves, check valves, pressure
regulators, flow meters, cavitating venturis, and propellant run tanks. In addition, new and improved methods to
accurately model the transient interaction between cryogenic fluid flow and immersed sensors that predict the
dynamic load on the sensors, frequency spectrum, heat transfer, and effect on the flow field are needed.

Propulsion System Exhaust Plume Flow Field Definition and Associated Plume Induced Environments
(PIE)

An accurate definition of a propulsion system exhaust plume flow field and its associated PIE are required to
support the design efforts necessary to safely and optimally accomplish many phases of any space flight mission
from sea level or simulated altitude testing of a propulsion system to landing on and returning from the Moon or
Mars. Accurately defined PIE result in increased safety, optimized design, and minimized costs associated with:

- Propulsion system and/or component testing of both the test article and test facility;
- Launch vehicle and associated launch facility during liftoff from the Earth, Moon, or Mars;
- Launch vehicle during the ascent portion of flight including staging, effects of separation motors, and
  associated pitch maneuvers;
- Effects of orbital maneuverings systems (including contamination) on associated vehicles and/or payloads
  and their contribution to space environments;
- Vehicles intended to land on and return from the surface of the Moon or Mars; and
- Effects of a vehicle propulsion system on the surfaces of the Moon and Mars including the contaminations
of those surfaces by plume constituents and associated propulsion system constituents.

In general, the current plume technology used to define a propulsion system exhaust plume flow field and its associated plume induced environments is far superior to that used in support of the original Space Shuttle design. However, further improvements of this technology are required:

- To reduce conservatism, in the current technology, allowing greater optimization of any vehicle and/or payload design while keeping in mind crew safety through all mission phases; and
- To support the efforts to fill current critical technology gaps (below). PIE areas of particular interest include: single engine and multi-engine plume flow field definition for all phases of any space flight mission, plume induced acoustic environments, plume induced radiative and convective ascent vehicle base heating, plume contamination, and direct and/or indirect plume impingement effects.

Current critical technology gaps in needed PIE capabilities include:

- An accurate analytical prediction tool to define convective ascent vehicle base heating for both single engine and multi-engine vehicle configurations;
- An accurate analytical prediction tool to define plume induced environments associated with advanced chemical, electrical and nuclear propulsion systems; and
- A validated, user friendly, free molecular flow model for defining plumes and plume induced environments for low density external environments that exist on orbit, as well as interplanetary and other planets.

**O2.03 Automated Collection and Transfer of Range Surveillance/Intrusion Data**

**Lead Center:** KSC

**Participating Center(s):** GSFC

Range surveillance is a primary focus of launch range safety and often a cost and schedule driver. Launch delays, due to the difficulty of verifying a cleared range, are common and will increase as spaceports are developed in new areas. Proposals are sought for sensors and communications technologies that expedite range clearance such as sonobouys; high altitude airships (HAAs) and related developments for thermal and gas pressure management, power systems, propulsion systems, and flight control; UAVs; use of commercial communication satellites for data transfer over the horizon; imaging through atmosphere and self learning/neural networks for pattern recognition.